Abstract— New instrumentation has been developed for non-contact, in vacuo measurements of the electron beam-induced surface voltage as a function of time and position for non-conductive spacecraft materials in a simulated space environment. The novel compact system uses two movable capacitive sensor electrodes to measure surface charge distributions on samples, using a non-contact method that has little effect on charge dissipation from sample. Design details, calibration and characterization measurements of the system are presented, with <1 V to >30 kV surface voltage range, <0.5 V calibration and characterization measurements of the system are presented, with <1 V to >30 kV, approximately 1.5 mm spatial resolution. Used in conjunction with the capabilities of an existing ultrahigh vacuum electron emission test chamber, the new instrumentation facilitates measurements of charge accumulation, bulk resistivity, effects of charge depletion and accumulation on yield measurements, electron induced electrostatic breakdown potentials, radiation induced conductivity effects, and the radial dispersion of surface voltage.

Three types of measurements of surface voltage for polyimide (Kapton HN) serve to illustrate the research capabilities of the new system: (i) accumulation using a pulsed electron beam, while periodically measuring the surface voltage; (ii) post charging, as deposited charge dissipated to a grounded substrate; and (iii), the evolution of spatial profile resulting from an incident Gaussian beam. Theoretical models for sample charging and discharge are outlined to predict the time, temperature, and electric field dependence of the sample's net surface voltage.

Index Terms—Surface charging, surface discharge, electrostatic discharge, materials testing, polyimide films, pulsed beams, test facilities, space environment interactions.

I. INTRODUCTION

Surface charging and subsequent electrostatic discharge due to interactions with the space environment is one of the primary concerns of spacecraft charging studies [1-4]. Laboratory measurements of the evolution of surface voltages and dissipation currents under simulated space conditions are the primary method used to determine the response of key materials to diverse incident fluxes.

This paper describes the design and use of a system to measure the accumulation and dissipation of surface charge on an insulator as a function of time and position in situ in a spacecraft charging vacuum test chamber. Surface charge is generated by incident fluxes that deposit charge and energy near the surface, and create secondary and backscattered electrons which are emitted from the material. The low charge mobility of insulators causes charge to accumulate where deposited, preventing even redistribution of charge and creating inhomogeneous local electric fields and potentials across the material; this drives charge transport and can lead to electrostatic breakdown. Deposited charge dissipates on relatively long time scales by charge transport through highly resistive materials to grounded substrates. The conductivity of the material is a key transport parameter in determining how deposited charge will redistribute across the spacecraft, how rapidly charge imbalances will dissipate, and what equilibrium potential will be established under given environmental conditions [1,5-6]. Hence, it is critical for reliable spacecraft charging models to use appropriate values of conductivity for thin film insulators to determine the correct charge distributions and charge decay times for the materials.

The bulk conductivity values of commonly used insulators have most often been determined using standard ASTM methods [7], with a parallel plate capacitor geometry and a voltage applied with electrodes (see Figure 1(a)). Similar tests have been done under vacuum conditions which are more analogous to space environments [8]. However, in many cases the charge storage method using surface voltage measurements [5,8-11] can measure lower conductivity values and is more similar to situations encountered in spacecraft charging [5,9,11]. Charge decay methods expose one side of the insulator in vacuum to incident charged particles, light or plasma, with a conductive electrode attached to the other side of the insulator. Data are obtained by capacitive coupling to measure both the resulting voltage on the open surface and emission of electrons from the exposed surface, as well as monitoring both conductive and displacement currents to the electrode (Figure 1(b)). In many space applications, the upper surface of the exposed insulator is floating and there is no electric field applied by electrodes but rather only electric field from self-charge.

The general design parameters of the system are set by the extent of the spacecraft charging problem [12]. A desired lower voltage range and voltage resolution is ≲ 1 V. This is...
estimated as ~10% of the electrostatic breakdown potential for thin film sample such as oxide layers or dielectric coating on the order of $10^8$ to $10^5$ m thick with typical electrostatic field strengths of $10^7$ to $10^8$ V/m and breakdown voltages of $10^4$ to $10^3$ V. A desired upper voltage range is $\geq 30$ kV [13]. This is the upper bound on incident electrons that most affect surface charging events [14], is also an upper bound on surface charging beyond which electrons penetrate far enough into materials that electron emission is minimal [15], and is the typical breakdown voltage for common ~100 µm thick blanket materials. Desired instrument response times can be estimated from dissipation times for low conductivity materials ($10^{-12}$ $(\Omega \cdot$ cm$)^{-1}$ to $10^{-20}$ $(\Omega \cdot$ cm$)^{-1}$) — with corresponding dissipation times of a few times $10^{-1}$ s to $10^3$ s — identified as problematic in spacecraft charging [16]. This suggests a response time on the order of 1 s is appropriate and a system stable over a few days would be required to see a few percent decay in the lowest conductivity materials [12]. Spatial resolution on the order of a few mm is also desirable, to facilitate monitoring lateral charge movement in dielectric samples on the length scale of a typical incident beam diameter.

The instrumentation is briefly described here, emphasizing how the sensor is incorporated into an existing detector. More extensive descriptions of the instrumentation and calibration — including the surface voltage probe (SVP) and electrostatic field transfer probe (EFTP) — are found in [12] and [17]. Three measurements are described to illustrate the research capabilities of the test system. Surface voltage measurements were made periodically during the electron beam charging process and as the surface voltage discharged to a grounded substrate after exposure. Analysis of the measured curves provides information about the material electron yields and bulk resistivity. The evolution of the spatial profile of the voltage across the sample surface was also measured by sweeping a small electrode across the surface.

### II. INSTRUMENTATION

Our novel surface voltage probe system is shown below to meet the general design guidelines for measurements most relevant to spacecraft charging issues. The response time of the probe and data acquisition system is fast enough to acquire data for lower resistivity materials such as low density polyethylene (LDPE), with a few seconds decay times. The long term stability and drift characterization required to measure at slow rates and take data over several days on materials that have a high resistivity like Kapton™ necessitate computer controlled data acquisition.

Design details, calibration and characterization measurements are presented for a system that meets the general design goals outlined above. The compact system uses two movable capacitive sensor electrodes (3 mm and 7 mm diameter) that can be swept across the sample using an in vacuo stepper motor to measure surface charge distributions on samples in situ, using non-contact electrostatic field probe methods that does not dissipate significant sample charge.

#### A. Overview of Electron Emission Test Chamber

The compact transfer probe design extends our measurement capabilities by allowing the surface voltage probe to fit within an existing hemispherical grid retarding field analyzer, so that surface voltages can be measured on samples tested using the extensive source flux and emission detection capabilities of an existing electron emission vacuum test chamber. An overview of the main electron emission test chamber is included to illustrate the full capabilities of the surface voltage test system. Further descriptions of this versatile ultrahigh vacuum (UHV) chamber are provided elsewhere [4,17-23] This chamber can simulate diverse space environments including controllable vacuum ($<10^{-10}$ to $10^{-3}$ Torr) and ambient neutral gases conditions, temperature ($<40$ to $>400$ K), as well as sources for a broad range of electron, ion and photon fluxes and energies.

Two primary electron sources provide monoenergetic electron beams ($\Delta E/E < 2 \times 10^4$) with electron energy ranges from $\sim 20$ eV to $\sim 30$ keV, beam spot full width at half maximum (FWHM) ranging from $\sim 50 \mu$m to $>100$ mm (depending on beam energy), and pulsing capabilities ranging from 10 ns to dc emission. Stable, uniform, well-characterized beam fluxes of 0.05 nA-cm$^{-2}$ to $>1$ µA-cm$^{-2}$ are possible from the electron guns. There are three ion guns with $<0.1$ to 5 keV monoenergetic sources for inert and reactive
The NIR-VIS-UV solar irradiance spectrum is simulated using a pair of pulsed, monochromated tungsten/halogen and deuterium RF powered continuum lamp sources produce focused (~0.5 cm diameter) radiation from 0.4 eV to 8.3 eV (150 nm to 2000 nm). Additional light sources include a Kr resonance lamp (10.3 eV), broadband Hg discharge and W-filament sources, and a variety of quasi-monochromatic NIR/VIS/UVA LED sources [4].

For conducting samples, electron guns are operated using a continuous, low-current beam of electrons, and dc-currents are measured with standard ammeters sensitive to \(\leq 10^{-13}\) A. The system at USU to measure electron emission from insulators uses a combination of methods to control the deposition and neutralization of charge. Typically, charge deposition is minimized by using a low current beam (~10-30 nA) focused on a sample area of ~7 mm\(^2\) that is delivered in short pulses of ~5 \(\mu\)sec (~150 fC or ~10\(^5\) electrons-mm\(^{-2}\) per pulse). For a typical ~100 \(\mu\)m thick dielectric sample, this amount of charge is estimated to change the surface potential by only 10\(^{-10}\) to 10\(^{-3}\) mV/pulse (positive) and requires ~500 pulses/sec to achieve an ~1 nA/cm\(^2\) dosage that typically causes discharge in space. The pulsed system uses custom detection electronics with fast (1-2 \(\mu\)s rise time) sensitive/low noise (10\(^4\) V/A /100 pA noise level) ammeters [20,21]. Detected current pulses are sent to a fast (1 GS/s) digital storage oscilloscope, equipped with resistive and induction sensors. Charge dissipation techniques include a custom low energy (~1-10 eV) electron flood gun for direct neutralization of positively charged surfaces between incident pulses [20,21,24] and use of visible and UV light sources for neutralization of negatively charged surfaces through the photoelectric effect. Sample heating to ~50-100 °C has also been used for dissipation of buried charge by thermally increasing the sample conductivity. Both DC and pulsed measurements and data retrieval are fully computer automated under LabVIEW™ control. A complete description of the DC-system and pulsed-system setups is provided in [18-21], along with additional insulator-yield and charging data.

**B. Detector Assembly**

A variety of detectors are available for measurements of single or simultaneous electron-, ion-, and photon-induced emission [18,20,21], including a standard Faraday cup detector, hemispherical analyzer, cylindrical mirror analyzer, and time of flight microchannel plate detector. Specifically, these allow us to measure total emitted electron (ion and photon) yield, backscattered/secondary yield, charge decay curves, and emission energy spectra [12].

The primary detector for emission studies is a custom hemispherical grid retarding field analyzer (HGRFA), with a retarding-field analyzer grid system for emitted-electron energy discrimination between back scattered electrons (energies >50 eV) and secondary electrons (energies <50 eV) (see Fig. 2). By ramping the grid (refer to labels K and L in Fig. 2) via a ramping grid bias, energy spectra of the emitted electrons can also be measured using this detector. The HGRFA features an
aperture and drift tube (M) for incident electron/ion admission and a fully-encasing hemispherical collector (J) for full capture of emitted electrons, that is particularly well suited and calibrated for absolute yield measurements [18,19,21]. The HGRFA detection system has been carefully calibrated to account for detector losses, allowing yield accuracies of better than 2% for conductor yields and 5% for insulator yields [4,17,18]. The HGRFA can be independently positioned in front of any sample (E) (see Fig. 2(a)). A low energy flood gun (N) and a variety of visible and UV LED light sources (O) are mounted on the HGFRA housing at near-normal incidence to provide neutralization of surface charging between pulses. A collimating lens mounted on the HRFRA and attached to a fiber optic cable and vacuum feedthrough allow external light sources to be used or a photospectrometer to analyze emitted light from the sample. The flood gun (N) also acts as a low energy (~1eV to 100 eV) focused electron source.

C. Sample Assembly

Samples (E) are typically mounted on (10.0 ± 0.1) mm diameter Cu cylinders [17]; sample up to 26 mm diameter can be accommodated. The Cu cylinders are mounted in sample blocks (F) on the sample carousel, and are electrically isolated. Electrical connection to the sample is made via one or more spring loaded pins (R) from the rear, allowing the current(s) to the sample to be monitored. The primary sample carousel is a right dodecagon that has eleven sample blocks that can be rotated in front of the various flux sources (see Figs. 2(c-e) in [17]). Typically, one sample block contains a photodiode, another a Faraday cup, and a third a Au sample as an electron emission standard. The sample carousel can be easily removed for rapid sample exchange and is mechanically positioned relative to the HGRFA face plate within ±0.5 mm. Ex situ tests showed no significant changes in the calibration factors for changes in probe to sample distances <5 mm.

The sample carousel is mounted on a cryogenic reservoir; it is electrically isolated using a ~75 μm thick Chotherm™ sheet that provides good thermal contact. Liquid nitrogen cooling allows sample temperatures maintained to within ±5 K. Temperatures ≥400 K can be achieved using resistive heating elements, held to within ± 1 K. The large thermal mass of the sample stage helps minimize temperature fluctuations. An alternative low-temperature sample stage has been developed for use with the HGRFA/SVP assembly [25]. The sample holder uses a closed cycle He cryostat to attain sample temperatures from ~2 K to >350 K, with ~0.5 K stability maintained by a standard PID temperature controller.

D. Surface Voltage Probe Design

The surface voltage probe (SVP) is a small device that fits within the HGRFA to measure the surface potential of a sample. Extensive details of the SVP system are given in [12] and [17] and a block diagram of the SVP system and electronics is shown in Figure 4 of [17]. Figure 3(c) shows the assembled SVP, which is <40 mm long and only ~21 mm wide, with a thickness of <3 mm. Two openings in the casing
of 7.0 mm (T) and 3.0 mm (U) diameter define the effective electrode areas. The casing is coated with colloidal graphite to minimize the production of secondary electrons by stray electrons inside the HGRFA (see Fig. 3(a)). There are two electrodes (U and T) on the sample side of the sensor ~500 µm above the sample surface, each kinematically positioned and electrically isolated by six 500 µm diameter sapphire spheres above and below the electrodes. The electrodes are Au plated to minimize surface contamination and allow a uniform charge density on the probe. Currents to the two electrodes, the Au disc, and the full SVP casing can be monitored independently because each are electrically isolated. The two voltage sensor plates are connected separately to external witness plates (X).

The SVP is mounted on a small sized (~25 mm x 11 mm diameter), ultrahigh vacuum-compatible stepper motor (Attocube Systems, Model ANR50res) (C). The microstepper controller (Model ANC200), with a resistive position encoder, provides rapid and extremely fine (<1 m° per step) positioning. The SVP can be positioned on either side of the sample providing an unobscured view for the incident beam and can be swept from side to side allowing either electrode to pass fully over the sample.

The SVP is a much smaller detector than commercial electrostatic field probes; this allows the SVP to be incorporated within the HGRFA (see Figs. 3(d) and 3(e)). The primary advantage positioning the SVP inside the HGRFA is that surface voltage measurements can be made rapidly, while the sample and HGRFA are accurately aligned with the incident beam. In addition, an electrically isolated 4.15 mm diameter Au disc (O) is mounted on the source side of the probe and can be swung into place above the sample in line with an incident beam, providing a Au electron emission calibration standard for the detector [21,26,27]. Further, the SVP in this position can act as a shield for the sample preventing any stray electrons or light from charging or discharging the sample. There is also a 360 µm diameter Faraday cup (S) in the source side of the probe that can be swept across the sample to center the beam on the sample.

E. Electrostatic Field Transfer Probe Design

The EFTP used here is based on Frederickson’s idea that a transfer probe can induce a surface voltage on an external witness plate proportional to the voltage on the sample and thus measure the sample voltage. However, since the EFTP still has no net charge (assuming the probe is fully isolated), an equal magnitude charge is found at the witness plate voltage with the external electrostatic field probe; this provides a measure of the zero offset \( V_{\text{offset}} \), which is the measured probe voltage for a grounded sample. A known voltage is then placed on a conducting sample, causing an equal magnitude and opposite polarity charge density to form on the voltage sensor plate. However, since the EFTP still has no net charge (assuming the probe is fully isolated), an equal magnitude charge is found at the opposite end of the EFTP. The charge density on the witness plate, \( \sigma_w \), is of the same polarity as the sample charge density, \( \sigma_s \), with magnitude of the witness plate charge density scaled by the ratio of the voltage sensor plate to witness plate capacitances, \( \sigma_w = (C_f / C_{wp}) \sigma_s \). The proportionality constant, \( C_f \), depends on the plate areas and separations, but can be determined directly by measuring the witness plate voltage with the external electrostatic field probe for a variety of applied sample voltages. Typical calibration errors for the system are shown in Fig. 5 of [17], and further details of the calibration process are given in [12]. Once calibrated, the EFTP can then be used to measure unknown surface voltages or charge densities of conducting or insulating samples.

In an ideal system, the probe has infinite resistance and zero capacitance coupling to ground. More correctly, one must consider the coupling of the EFTP to ground, including both the capacitance of the wire and probes and the leakage resistance to ground through the feedthrough, wire insulation, and probe mounts [30]. An expression for the time-dependant voltage on the sample, \( V_s \),

III. CALIBRATION AND MEASUREMENTS

A. Measurement and Calibration Principles

To accurately measure a surface voltage with the EFTP, the sample plate and witness plate are positioned adjacent to grounded surfaces and the EFTP is grounded. This assures that there is no net charge on the EFTP and that the charge density is zero on both plates. The EFTP is then disconnected from ground and the witness plate voltage is measured with the electrostatic field probe; this provides a measure of the zero offset \( V_{\text{offset}} \), which is the measured probe voltage for a grounded sample. A known voltage is then placed on a conducting sample, causing an equal magnitude and opposite polarity charge density to form on the voltage sensor plate. However, since the EFTP still has no net charge (assuming the probe is fully isolated), an equal magnitude charge is found at the opposite end of the EFTP. The charge density on the witness plate, \( \sigma_w \), is then of the same polarity as the sample charge density, \( \sigma_s \), with magnitude of the witness plate charge density scaled by the ratio of the voltage sensor plate to witness plate capacitances, \( \sigma_w = (C_f / C_{wp}) \sigma_s \). The proportionality constant, \( C_f \), depends on the plate areas and separations, but can be determined directly by measuring the witness plate voltage with the external electrostatic field probe for a variety of applied sample voltages. Typical calibration errors for the system are shown in Fig. 5 of [17], and further details of the calibration process are given in [12]. Once calibrated, the EFTP can then be used to measure unknown surface voltages or charge densities of conducting or insulating samples.
\[ V_s = \left( \frac{C_W + C_I}{C_I} \right) V_{\text{probe}} e^{-t/\tau} \quad \text{with} \quad \tau = R C_W \quad (1) \]

uses an exponential decay of the initial probe voltage \( V_{\text{probe}} \) with time as charge leaks into (or from) the EFTP, with an RC time constant, \( \tau \) (see details in [12] and [17]). The three largest sources of a resistance for \( R \) are the leakage through air of the witness plate to the EFTP (~4×10^{14} \Omega), the electrical isolation of the electrodes to the probe body through the sapphire spheres (~3×10^{14} \Omega), and the vacuum feedthrough to ground (~1×10^{14} \Omega). The highest sources of capacitance of the probe, \( C_s \), are the capacitive of the feedthrough (~12 pF) and the capacitive of the wire (~6 pF).

Another reason for preferring the EFTP arrangement relates to electron emission from insulators [5]. Electron beam charging of the samples produces an electric field at the surface of the sample that can drive electrons out of the surface. While penetrating into the insulator, the high-energy electrons excite electrons and holes into trapping states and into mobile states located in the region between the sample surface and the maximum depth of penetration. Such trapped charge provides the charge to be later emitted from the surface. This effect is sometimes termed the Malter effect [28,32]. An in situ electrostatic field probe can collect these delayed emitted electrons, thereby altering the net charge on the electrostatic field probe and modifying the voltage reading with time. The same modification of the net charge on the EFTP can occur for ex situ electrostatic field probes. However, by knowing the sample surface plate to voltage sensor plate capacitance, \( C_s \), the rate of voltage change on the voltage sensor plate provides a direct, sensitive determination of the electron currents leaving the sample surface. [12] and [17] describe a calibration method to account for this effect.

B. Calibration

To determine the calibration factor of the EFTP, measurements were made of the probe voltage for a series of known sample voltages. The large electrode has a repeatable calibration factor of \( CF=1084.5 \pm 0.5 \) V/V, over a range of applied voltages >1000 V. It is good practice to determine the calibration factor for each set of experiments as well, as there is some small variation due to specific sample and sensor conditions and sample-to-sensor separation. The calibration factor was approximately proportional to this separation, as expected for the parallel plate geometry; however, this separation was quick reproducible for a given sample and variations from one sample to the next were accounted for by measuring a calibration factor for each sample. Tests also indicated that an accurate surface voltage measurement could be made in ~500 ms, as limited by the time constant of the EFTP (~100 ms), the response time of electrostatic field probe (<5 ms), and data acquisition time.

The probe offset voltage (typically on the order of a few mV) and voltage drift with time were found to differ for each test and must be measured for each test sequence by performing an applied voltage calibration run. To calibrate the EFTP drift due to leakage, a constant voltage was applied to the sample and the probe voltage was monitored with time over ~2 hr. At a zero applied voltage, the probe voltage was found to change almost linearly with time over early times at a rate of \( (V_{\text{drift}}/t_d) = 280 \) µV/s. Measurements made for nonzero applied voltages produced very similar drift rates. Without correcting for voltage drift, there would be a ~0.5 V error in measured surface voltage, comparable to the instrument resolution, in ~12 s. After correcting for a linear drift, measurements can be taken for >4 hr with <20 V error.

Hodges provides a much more detailed discussion of the calibration that becomes relevant for operation of the SVP requiring higher precision or longer times between recalibration [12]. His calibration extends the linear approximation for drift in Eq. 2 to an exponential correction as expressed in Eq. 1 and additional corrections for the drift in the detector electronics and exponential drifts in in situ and voltage of the sample voltage.

Combining the results of the calibration tests, the measured probe voltage is related to the actual surface voltage through a linear approximation to Eq. 1 as:

\[ V_s(t) = CF \left[ V_{\text{probe}} \left( 1 + \frac{t-v}{v_{c}} \right) - V_{\text{offset}} + \frac{V_{\text{drift}}}{t_d} (t-t_d) \right] \quad (2) \]

where \((t-t_d)\) and \((t-v)\) are the elapsed time since recalibration of the probe to a grounded surface and the time that the probe has been positioned over a sample, respectively. For times \( \leq 150 \) s voltage drift is negligible (i.e., \( \beta=0 \)) and a linear approximation for the temporal drift introduced errors less than other sources of error. The EFTP and SVP assembly was sensitive to a surface voltage of <1 V with a resolution of ~0.5 V. Surface voltages up to ±12 kV could be measured with the Monroe probe. Much higher voltages (in principle up to ±30 kV) could be measured with a Trek electrostatic field probe. A modest voltage drift rate was observed in the sample voltage of <3 mV/sec. Without correction for drift, surface voltages can be measured for short periods of time—long enough for accurate surface sweeps—between recalibration of the probe. With a linear voltage drift correction, surface voltages can be measured to high accuracy for periods >4 hr between probe recalibration.

Data were acquired and processed using an automated Labview program. The SVP data are typically sampled at 1 kHz for 1 s intervals; averages and standard deviations are retained. Figure 4 shows a typical timing diagram for data acquisition. 10 s of data with the SVP positioned over a grounded plane are acquired before and after a 10 s interval of data acquired with the SVP positioned over the sample; \( V_{\text{offset}} \) and \( (V_{\text{drift}}/t_d) \) are determined through a linear fit to the grounded data, for offset and drift corrections using Eq. 2. For charge accumulation experiments, the SVP is then retracted.
and the electron beam is un-blanked for different lengths of
time from 10 s to 120 s. 1 s wait times were included after
SVP movement to allow dissipation of electronic noise.

IV. APPLICATIONS

Three types of measurements have been made on the
prototypical polymeric spacecraft material, polyimide, to
illustrate the research capabilities of the new system [12]. The
polyimide sample was a 25 µm thick film of Kapton HN™
from Dupont, with a relative dielectric constant, $\varepsilon_r$, of 3.5 and
a total electron yield of $Y \approx 0.32$ at 5000 eV. First, using a
pulsed electron beam, periodic measurements of the surface
voltage were made (see Fig. 5 (a)). A total dose of 3 µC-cm$^{-2}$
was delivered over ~60 min. Charge pulse durations were
increased from 10 s to 30 s at approximately 340 s of injected
charge and then again from 30 s to 120 s at approximately 890
s of injected charge. Second, post charging measurements of
the surface voltage were conducted, as deposited charge
dissipated to a grounded substrate over a ~50 hr period (see
Fig 5(b)). Finally, surface voltage spatial profile
measurements were made twice during the charging process
and then periodically as the sample discharged to a grounded
substrate after exposure (see Fig. 5(c)).

Theoretical models for sample charging and discharge are
presented, based on dynamic bulk charge transport equations
developed for electron charge carriers to predict the time,
temperature, and electric field dependence of the sample net
surface voltage [10]. The model includes electron drift,
diffusion, and displacement currents and makes direct ties to
the interactions between injected electrons, which are trapped
in localized states, and the magnitude and energy dependence
of the density of those localized trap states within the gap; the
carrier mobility, and the carrier trapping and de-trapping rates
are then evaluated using the model. An overview of the
models is provided in [33], with more details in [10] and [12].

For the experimental conditions considered here, the
generalized time-dependant conductivity for non-Ohmic
conductivity in highly disordered insulating materials [10] is
restricted to: (i) times less than the transit time or $t < (\varepsilon_0\varepsilon_r/\sigma_{DC})$; (ii) polarization has come to equilibrium before
the first decay point is acquired; (iii) persistent radiation
induce conductivity (RIC) has dissipated. This leaves

$$\sigma(t) = \sigma_0 \left\{ 1 + \frac{\sigma_{\text{diffusion}}}{\sigma_0} t^1 + \frac{\sigma_{\text{dispersive}}}{\sigma_0} t^{(1-\alpha)} \right\}$$

where $\sigma_0$, $\sigma_{\text{diffusion}}$, and $\sigma_{\text{dispersive}}$ represent the long-time quasi-equilibrium conductivity, the propagation of the centroid of
the trapped space charge distribution through diffusion or hopping, and the spreading of the distribution in space, respectively. The dispersion parameter, \( \alpha \), is a measure of the width of the energy distribution of trap state below the conduction band edge. These four parameters are determined from simultaneous fits to the charge and discharge data below.

We found \( \sigma_0 \sim 5 \times 10^{20} \text{ (} \Omega \text{-cm)}^{-1} \), which is close to independent room temperature dark current conductivity of \( 1.5 \times 10^{19} \text{ (} \Omega \text{-cm)}^{-1} \) measured with a constant voltage charge injection method [8] and \( 1.7 \times 10^{19} \text{ (} \Omega \text{-cm)}^{-1} \) measured with a charge storage method [28]. \( \sigma_0^{\text{diffusion}} \) and \( \sigma_0^{\text{dispersive}} \) were found to be \( -6.4 \times 10^{-15} \text{ (} \Omega \text{-cm)}^{-1} \) and \( -7.9 \times 10^{-15} \text{ (} \Omega \text{-cm)}^{-1} \), respectively. These dominate the conductivity at early times. The value \( \alpha = 0.55 \) from the fit is reasonable for Kapton.

A. Charge Decay

The discharge curve is shown in Figure 5(b). These data are fit with a simple form of the voltage relation that follows from the value estimated by [34].

To estimate the spatial resolution of the system, the spatial profile of a 10.0 mm diameter conducting Au sample (E) uniformly charged to 1000 V was measured by sweeping the 7.0 mm diameter Au voltage sensor electrode (T) over the sample at 1.3 mm increments. A convolution of the circular electrode, estimated as a circular step function, over this charged conductor (also estimated as a circular step function) provides an estimation of the voltage profile. The spatial resolution for the larger diameter probe after deconvolution is estimated to be 1 mm to 2 mm.

An experiment was performed to measure the time evolution of the non-uniform radial charge profile of a sample irradiated with an electron beam. Measurements are shown in Fig. 5(c) of the charge distribution on a 10.0 mm diameter polyimide sample produced by \( 60 \text{ s} \) of cumulative irradiation using a well-characterized, focused, Gaussian electron beam with a measured full width at half maximum of 5.6 mm [44]. The initial charge profile \( 145 \text{ s} \) after charging, along with six subsequent profiles, are shown. After \( 60 \text{ s} \) of cumulative irradiation, the sample charge reached a maximum peak voltage of \( 1456 \pm 5 \) V. The peak voltage values decayed over the following 24 hr period as shown in Fig. 5(d); the surface voltage decay was similar to that observed in the discharge experiment described above.

Initially, the deconvoluted voltage profile was equal to a Gaussian profile with a FWHM of 6.7 mm, approximately 1.1 mm wider than the incident electron beams measured FWHM. Over time the radial charge profile widths observed in Fig. 5(c) did not change, indicating that that charge dissipation through, not across, the sample was the dominate process and that no measurable radial diffusion occurred after the initial profile was measured \( 200 \text{ s} \) after deposition began. These results suggest that if any diffusion occurred, the majority of the radial diffusion happened during the first \( \approx 200 \text{ s} \) when the charging of the material was taking place and radiation induced conductivity (RIC) was active in the thin surface layer, where the beam penetrated. The equilibrium value of...
Ric while the 5 keV, 1.1 nA/cm² beam is on is estimated as $\sigma_{RIC} \approx 3 \times 10^{-14}$ (Ω-cm)$^{-2}$, using a RIC coefficient of $-6 \times 10^{-18}$ (Ω-cm-rad/s)$^2$ [45] and a dose rate of $\sim 8 \times 10^3$ rad/s up to a range of $\sim 0.5$ µm [15]. The average RIC over the time before the first profile is $\sim 40\%$ of the beam on value, after accounting for the beam duty cycle and the hyperbolic decay of persistent RIC with a decay constant of $\sim 220$ s [36]. A crude estimate of the mean lateral velocity of the charge FWM is ($>30$ kV theoretical) with a resolution of $<0.5$ V. The beam duty cycle and the hyperbolic decay of persistent RIC probe (SVP) was demonstrated to have $\sim 1$ V to $>1$ kV range extent of the spacecraft charging problem. The surface voltage profile is taken; this approximate value is $\sim 60\%$ of the additional measurements and more realistic calculations are required to confirm this hypothesis.

V. CONCLUSION

This paper described the design of a versatile high sensitivity surface voltage probe system. We have verified that the new system meets the general design parameters as set by the extent of the spacecraft charging problem. The surface voltage probe (SVP) was demonstrated to have $\sim 1$ V to $>1$ kV range ($>30$ kV theoretical) with a resolution of $<0.5$ V. The response time was found to be about $\sim 1$ s and the spatial resolution was $\leq 1.5$ mm. The instrument was successfully applied to three different types of experiments: electron-induced charging, surface discharge, and measurements of spatial profiles and surface diffusion. These experiments were shown to be modeled well by standard theory and consistent with literature values of the materials parameters. The “potential” for the instrument for important applications in spacecraft charging has clearly been demonstrated.

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